

ABSTRACT OF THE DISCLOSURE

An assembly for a gas turbine engine, the assembly comprising a combustor and a vane assembly downstream from the combustor. The vane assembly includes at least one airfoil radially extending between an inner and outer platform defining an annular gas path therebetween. At least the outer platform forms a first sliding joint connection with an adjacent outer combustor wall such that radial sealing between the outer platform and the outer combustor wall is provided at engine operating temperature, while permitting relative axial displacement therebetween.